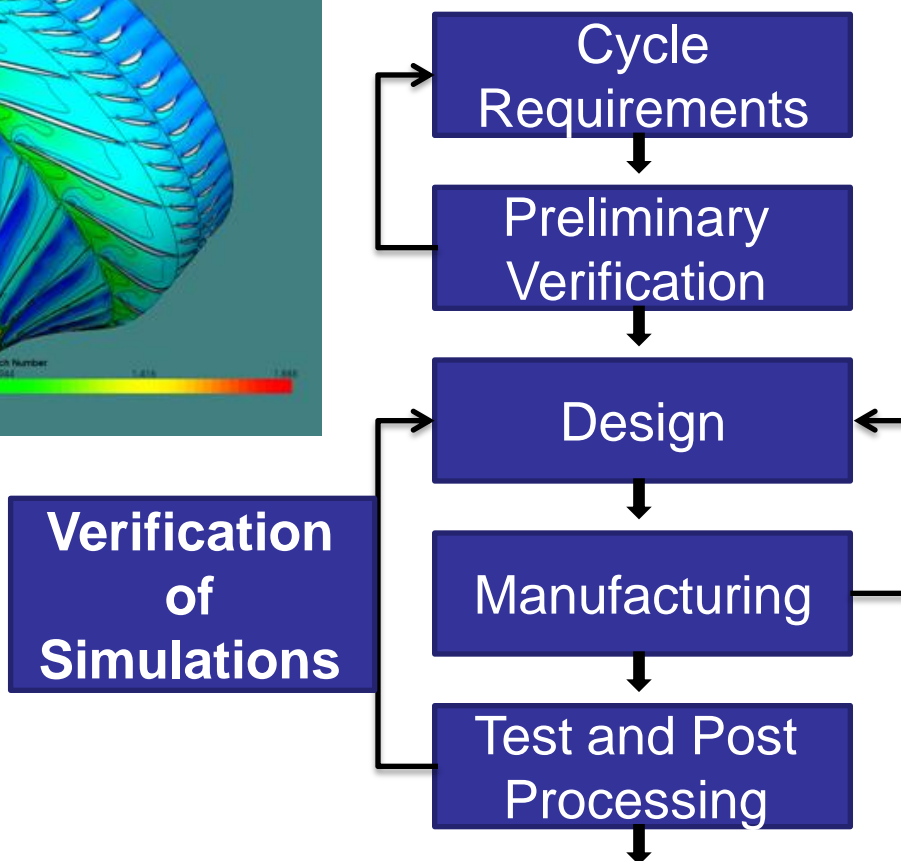
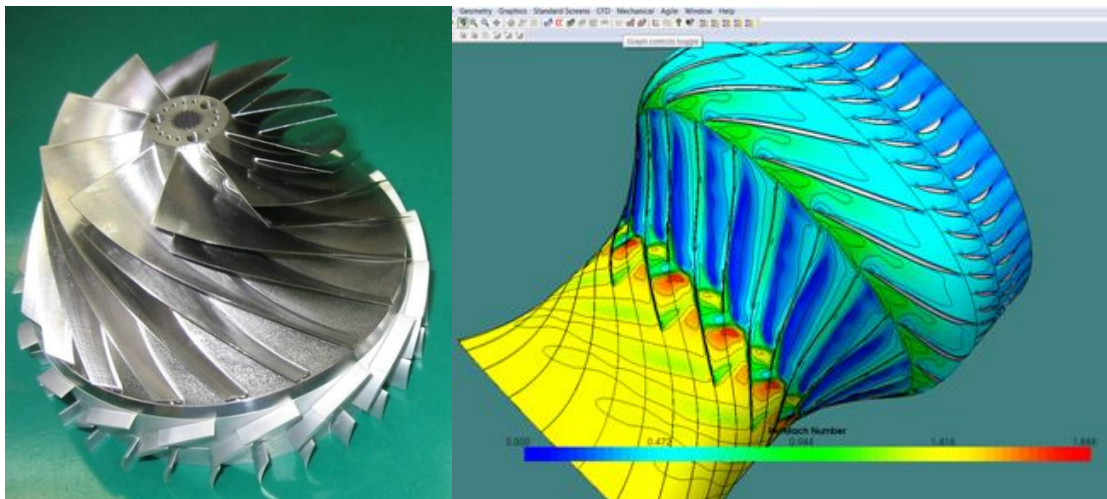


Compressor Development Process – Verification of Simulations

**Konstantin Rosenberg
Jet Propulsion Department,
Rafael Advanced Defense Systems, Haifa, Israel**

**The 15th Israeli Symposium on Jet Engines and Gas Turbines
*Technion, Haifa , November 17 ,2016***

Compressor Development Process

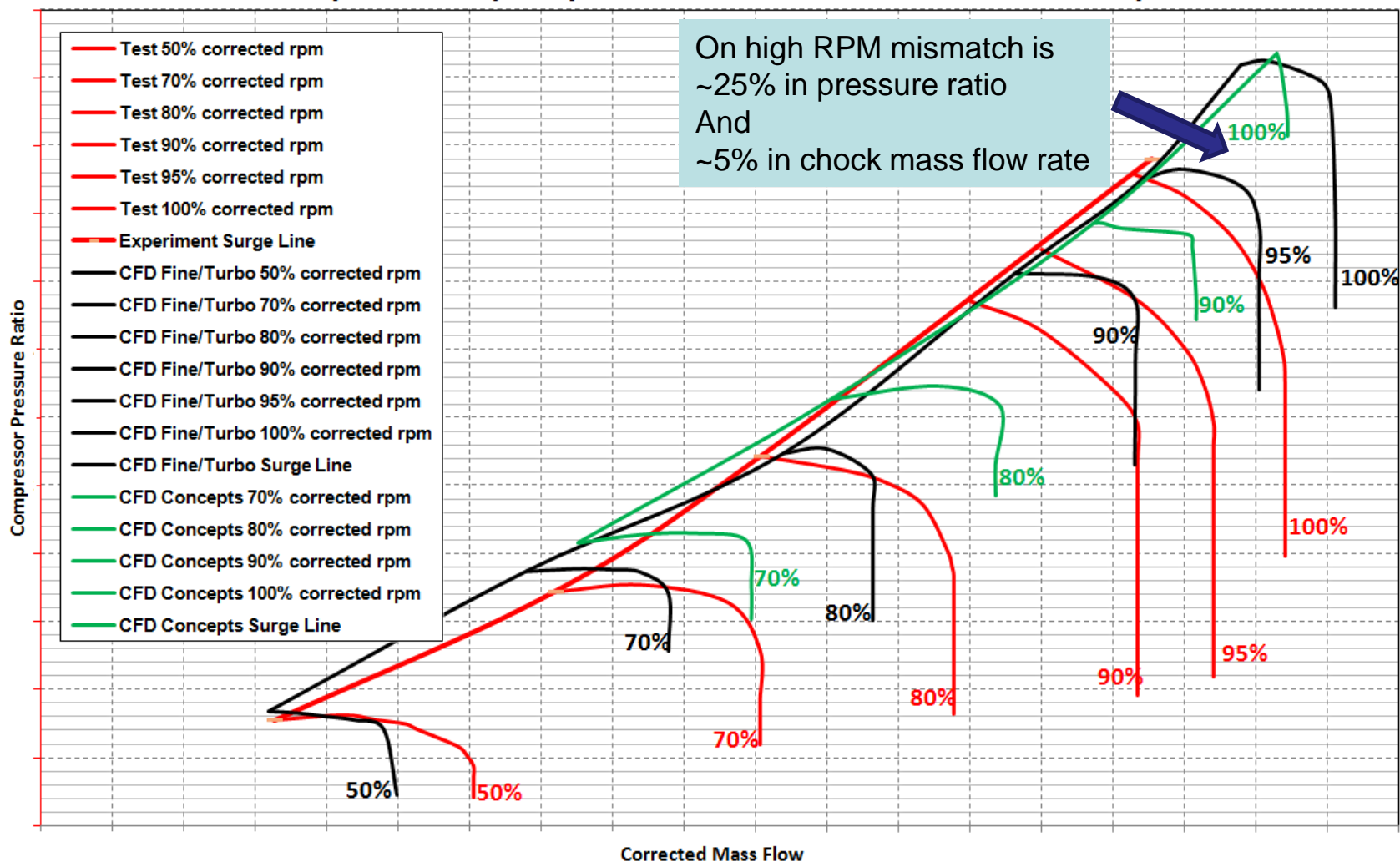


Post processing and Verification of Simulations For Mix Flow Compressor

- 1) Experimental mapping of compressor prototype.
- 2) Discovered mismatch between predicted and measured characteristics map of the compressor.
- 3) The compressor test bench was validated with a proved compressor design
- 4) Verification of simulation:
 - a) Mapping causes for mismatch
 - b) Running series of simulations after improving mismatch causes

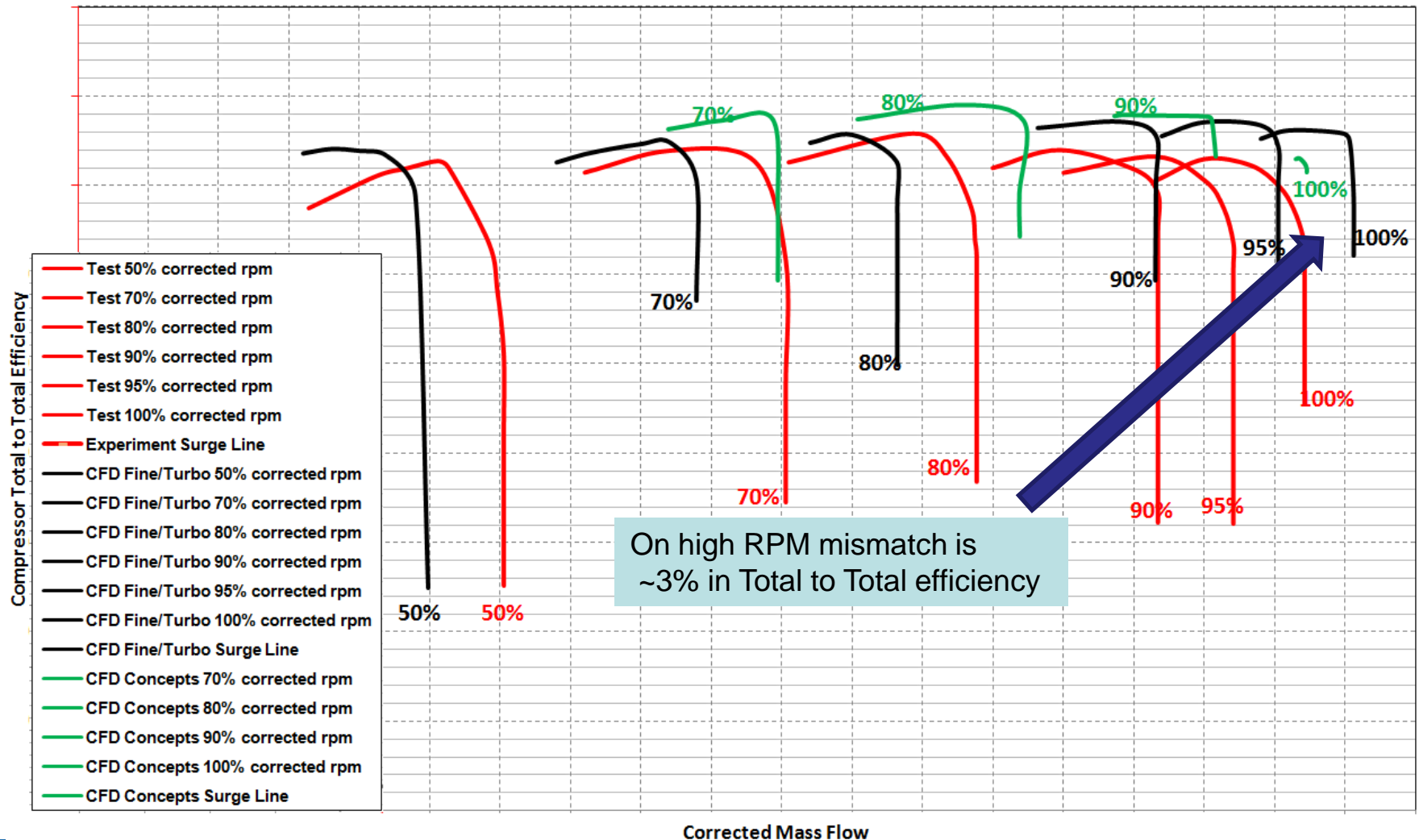
Comparison between Test results and CFD predictions

Compressor Map - Experimental Results Vs. Numeca & Concepts



Comparison between Test results and CFD predictions

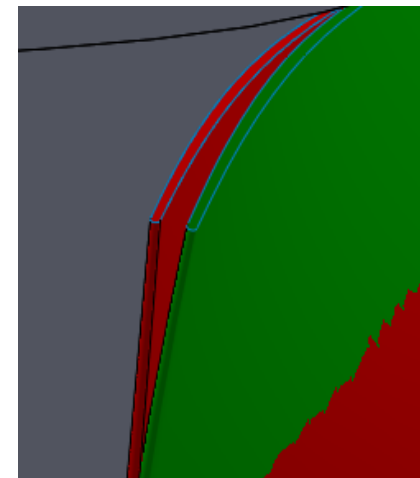
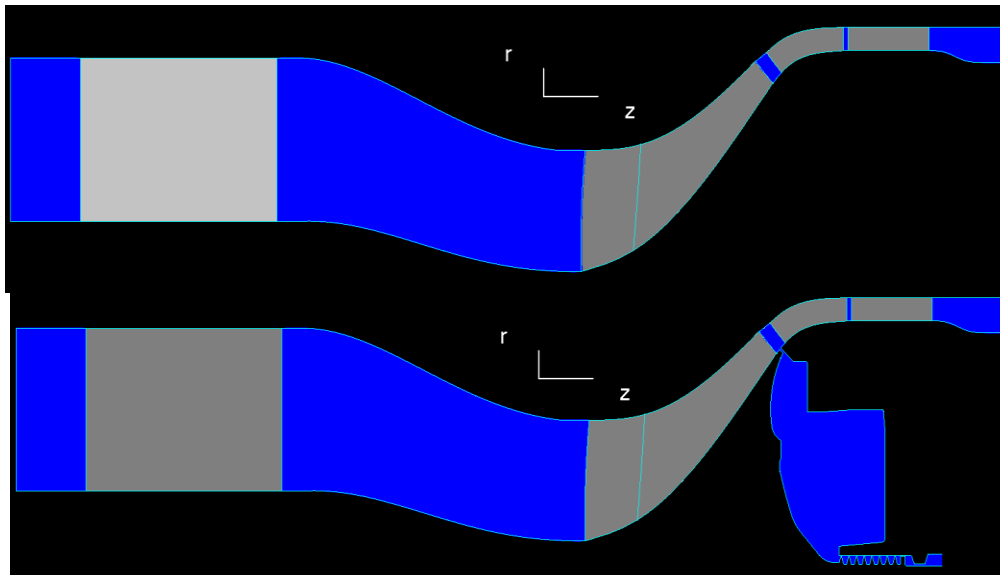
Compressor Map - Experimental Results Vs. Numeca & Concepts





Reasons for Mismatch Between Test Results and CFD Predictions

- 1) Blade deformations due to centrifugal forces had not been encountered.
- 2) Radial clearance in test was 0.5 instead of 0.35.
- 3) Impeller backward cavity had not been encountered.
- 4) Steady state Solution is probably not enough accurate in case of narrow distance between impeller and first diffuser.



Verification Approach

- 1) Geometry must match the real one as much as possible (cavities, seals, deformations and etc').
- 2) Clustering of the grid in problematic areas .
- 3) Testing various approaches of mixing plane. Not all mixing planes good enough for super sonic flow.
- 4) Testing various turbulence model.
- 5) Testing various discretization scheme (central differences or upwind).
- 6) Harmonic calculation



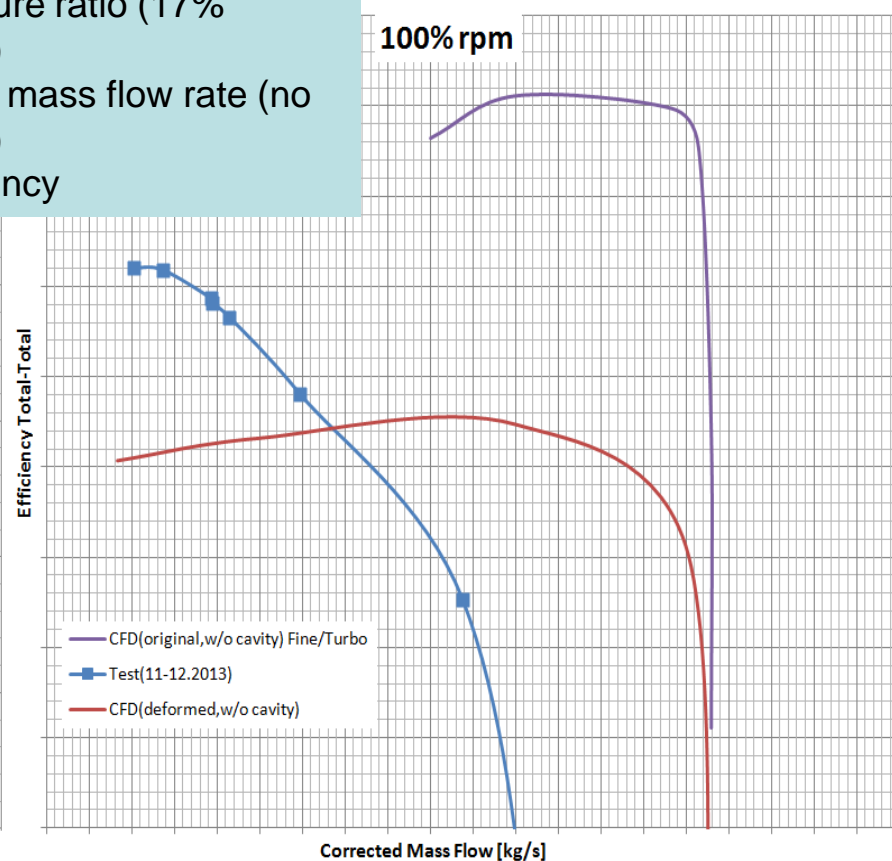
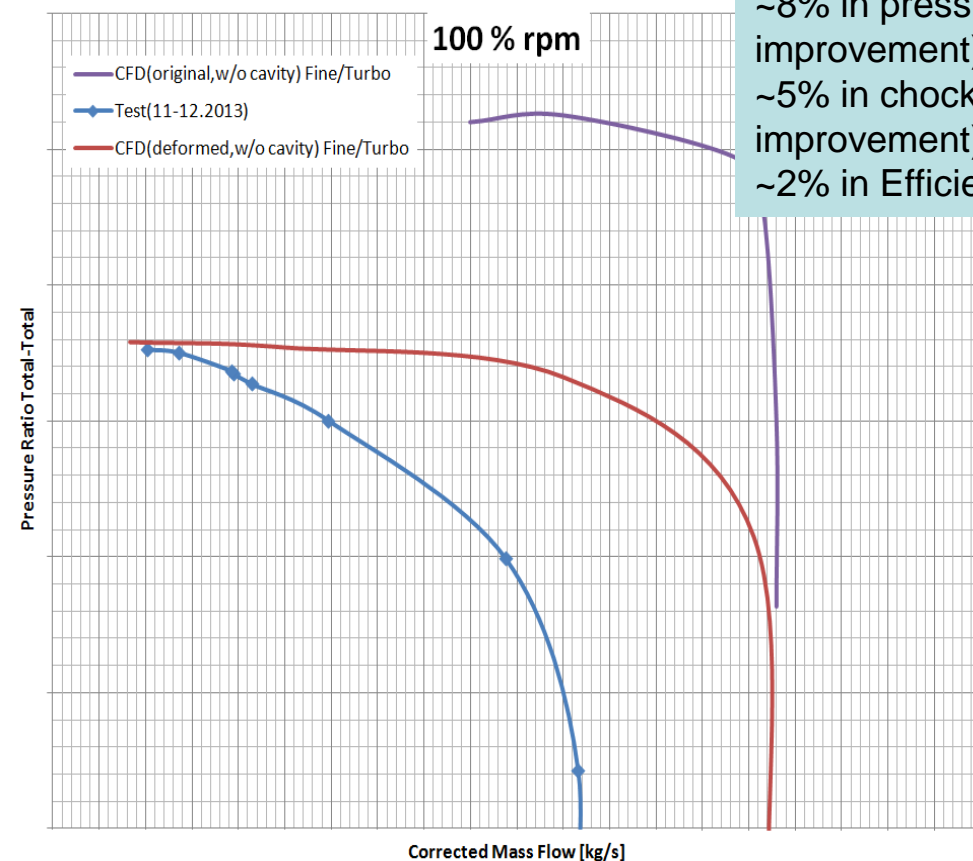
Numerical Approaches Used In Project

- 1) Full non-matching interface instead of rotor-stator interface between 1st and 2nd diffusers.
- 2) Non reflective 1D rotor-stator interface between impeller and 1st diffuser.
- 3) Spalart-Almaras turbulence model.
- 4) Central differences were used for discretization scheme.
- 5) 3 level multigrid.
- 6) Full non-matching interface between impeller hub and impeller rear cavity.



Improving CFD Prediction by Taking into Account Blade Deformations Due to Centrifugal Forces

Mismatch after fixed:
~8% in pressure ratio (17% improvement)
~5% in chock mass flow rate (no improvement)
~2% in Efficiency



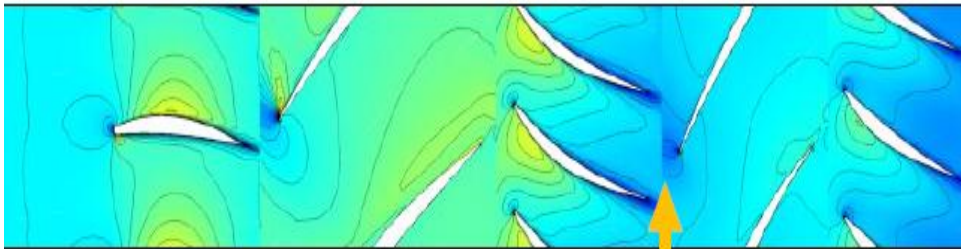


Future Steps to Improve CFD Prediction

Numeca Unsteady Treatment NLH Introduction

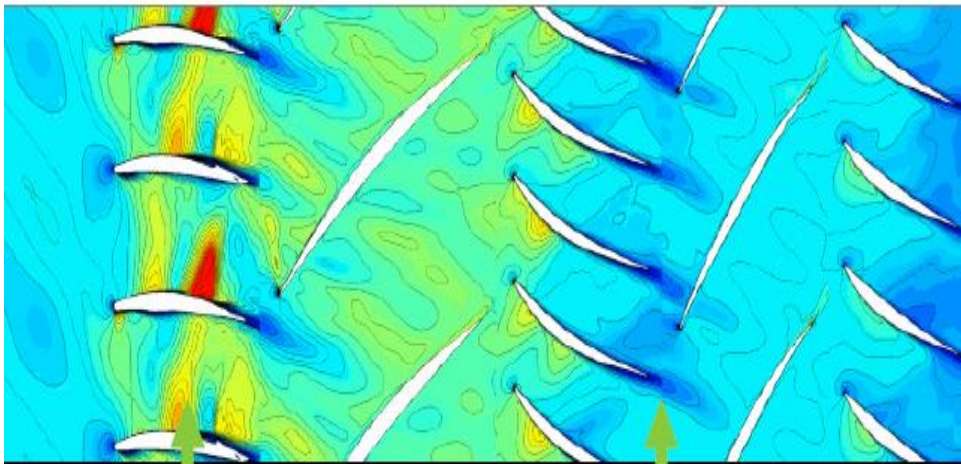
Steady

Absolute Mach Number



NLH

Mach number continuity is averaged at Rotor/Stator Interface



Unsteady shock due to Blade 1 potential

Mach number continuity is respected and not averaged

- Including impeller rear cavity in simulation.
- NLH simulation for better prediction of the first stator performance.
- Verify simulation tools for future design improvements.



Summary

- 1) Geometry must match the real one as much as possible (cavities, seals, deformations and etc').
- 2) Testing various numerical approaches.
- 3) Harmonic calculations to understand unsteady physics.



Thank you !